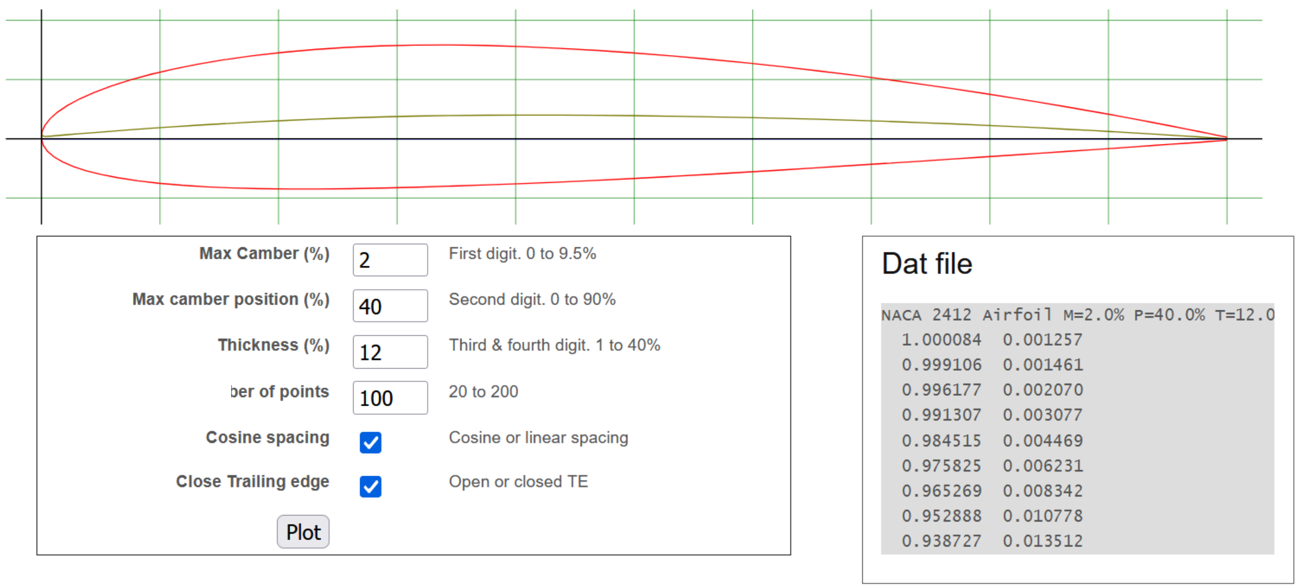
(KEY CONCEPT) [C-1-3]

NACA 4 digit airfoil generator (NACA 2412 AIRFOIL) http://airfoiltools.com



Max Camber (%) 2 First digit. 0 to 9.5%

Max camber position (%) 40 Second digit. 0 to 90%

Thickness (%) 12 Third & fourth digit. 1 to 40%

ber of points 100 20 to 200

Cosine spacing ☑ Cosine or linear spacing

Close Trailing edge ☑ Open or closed TE

Dat file

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NACA 2412 Airfoil M=2.0% P=40.0% T=12.0
  1.000084  0.001257
  0.999106  0.001461
  0.996177  0.002070
  0.991307  0.003077
  0.984515  0.004469
  0.975825  0.006231
  0.965269  0.008342
  0.952888  0.010778
  0.938727  0.013512
```

NACA 4-Digit Series: NACA X ① X ② XX ③

① One digit describing maximum camber (in % of chord).
② One digit describing the distance to the maximum camber location measured from the leading edge (in ×10% of chord).
③ Two digits describing maximum thickness of the airfoil (in % of chord).

NACA 5-Digit Series: NACA X ① XX ② XX ③

① One digit, when multiplied by 1.5, gives the lift coefficient in 1/10.
② Two digits, when divided by 2, describe the distance to the maximum camber location measured from the leading edge in 1/10 of chord.
③ Two digits describing the maximum thickness of the airfoil in % of chord.

NACA conventional airfoils (source: airfoiltools.com)