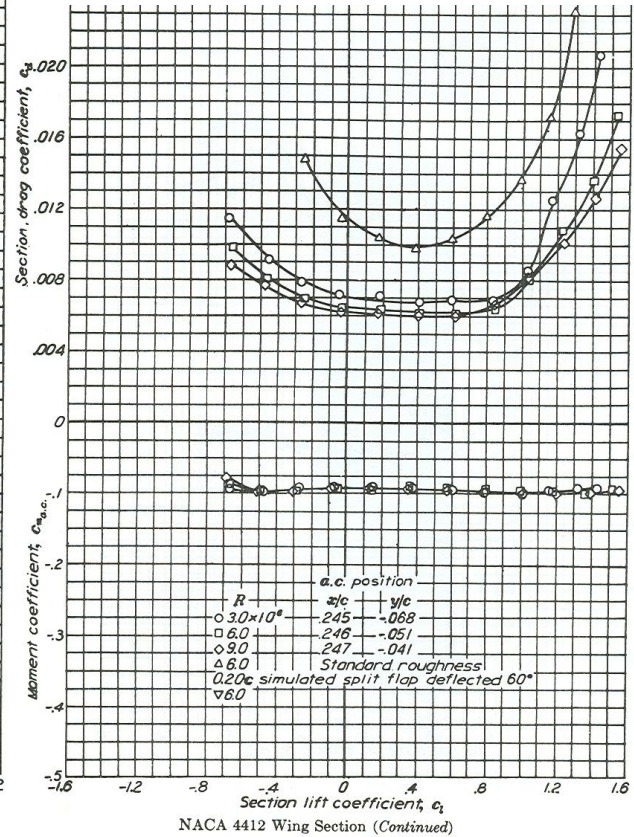
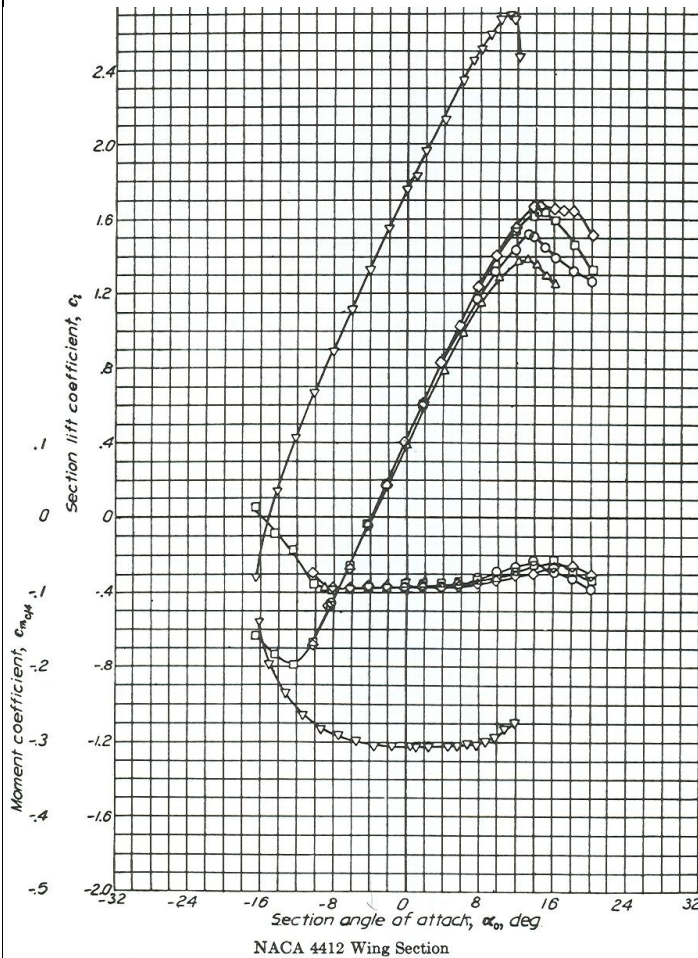


(EXAMPLE) [D-1-1]

Consider a flying wing (such as Northrop B-2) with a wing planform area of 200 m^2 , aspect ratio of 10, span efficiency factor of 0.95, and NACA 4412 airfoil (note: all hypothetical). The weight of the airplane is $7.5 \times 10^5 \text{ N}$. For the level-flight condition (assume $L = W$ and minimum profile drag condition) with 0.5 Mach at 10 km altitude, estimate the total drag on the aircraft.



Lined area for notes, consisting of multiple horizontal dashed lines.