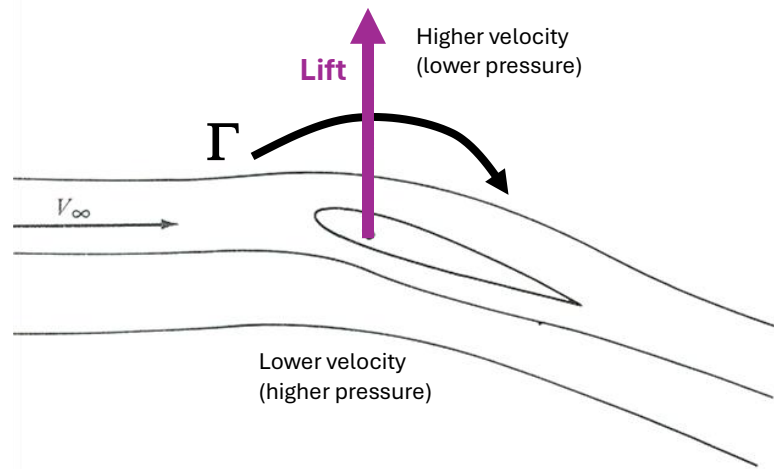


(EXAMPLE) [B-4-2]



Consider an airplane with a NACA airfoil (with chord length of 1 m). The airplane's airspeed is 50 m/s at the altitude of 4 km. The corresponding lift coefficient of the airfoil is $c_l = 0.8$.

- (a) Determine the lift per unit span.
- (b) Calculate the corresponding circulation around the airfoil.

Lined area for notes, consisting of multiple horizontal dashed lines.