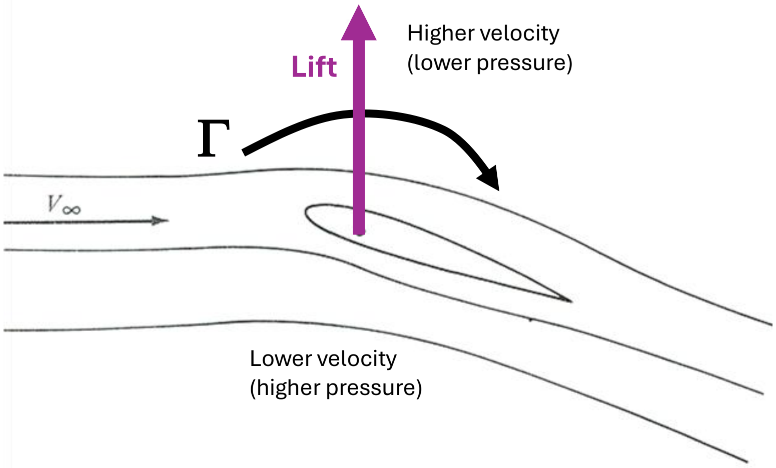
(EXAMPLE) [B-4-2]



Consider an airplane with a NACA airfoil (with chord length of 1 m). The airplane's airspeed is 50 m/s at the altitude of 4 km. The corresponding lift coefficient of the airfoil is c_l = 0.8.

(a) Determine the lift per unit span.

(b) Calculate the corresponding circulation around the airfoil.